

# Benchmark Supercritical Wing (BSCW)



Fluid Dynamics

Structural Mechanics

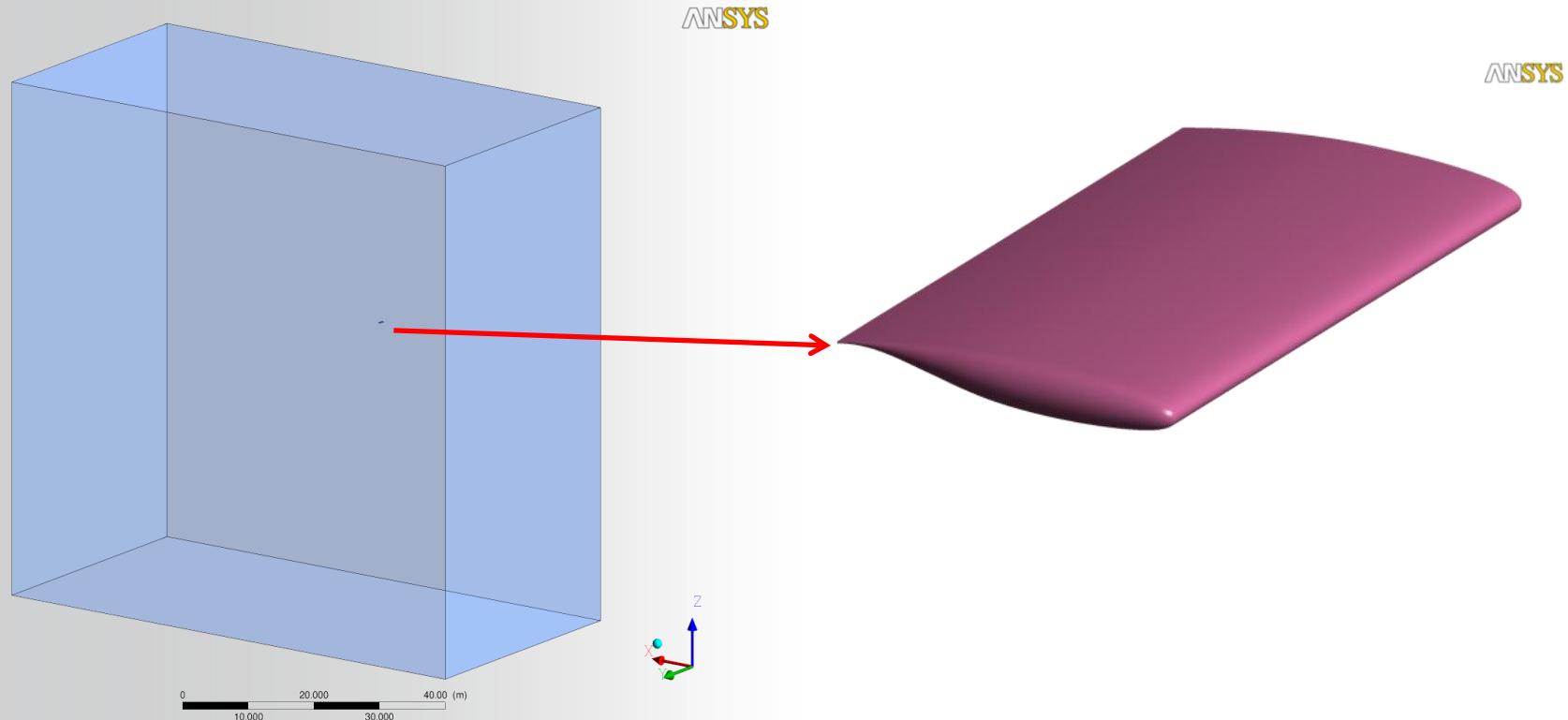
Electromagnetics

Systems and Multiphysics

**Angela Lestari, Thorsten Hansen**  
**ANSYS**

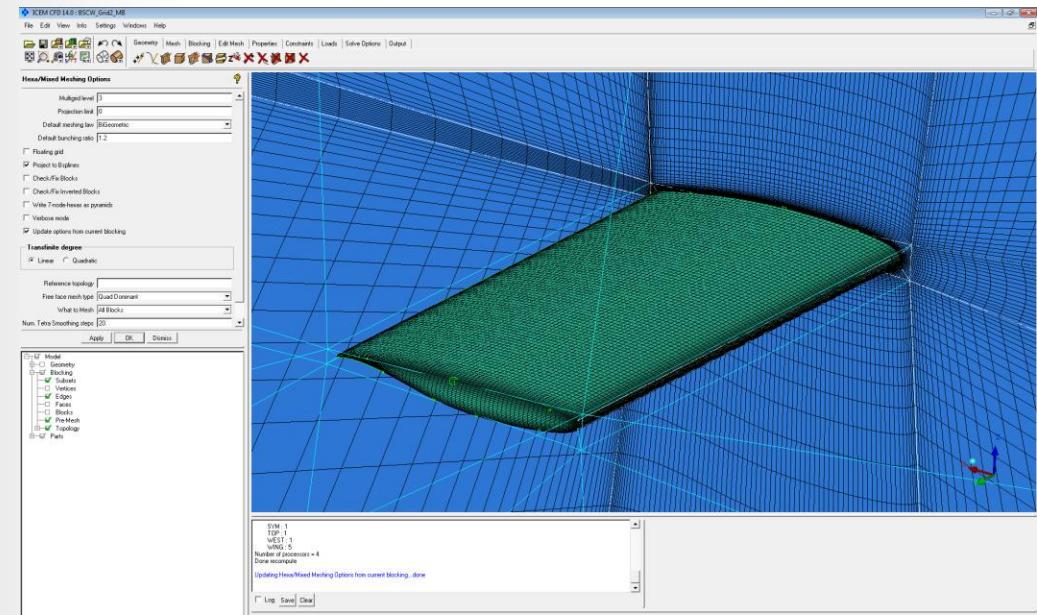
# Computational Domain

- $C_{ref} = 16.0$  inches (406.4 mm)
- $100 * C_{ref}$  in all directions



# Grid Information

- ANSYS ICEM CFD 14
- Hexahedral elements
- Scalable grids
  - Consistent mesh quality upon grid refinement
- Multigrid
  - levels = 3

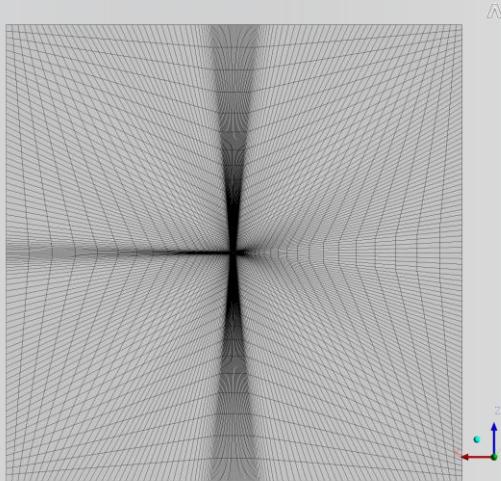


# Grid Information

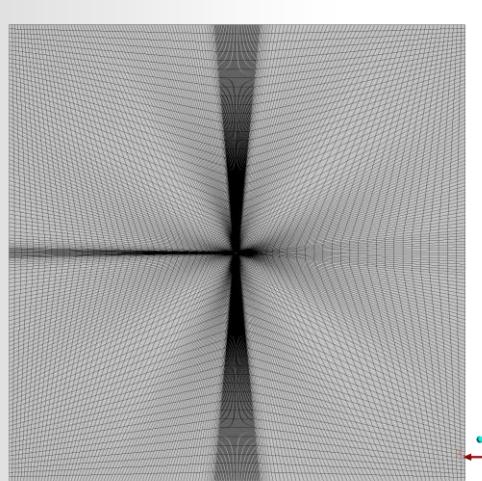
	Grid 1	Grid 2	Grid 3
<b>Number of nodes</b>	<b>1,488,809</b>	<b>5,025,665</b>	<b>13,926,249</b>
<b>Number of elements</b>	<b>1,448,960</b>	<b>4,935,680</b>	<b>13,747,200</b>
<b>Minimum grid angle</b>	<b>31.41°</b>	<b>32.43°</b>	<b>29.07°</b>
<b>Maximum aspect ratio</b>	<b>23,250</b>	<b>19,356</b>	<b>16,387</b>
<b>First grid node @ Wall, inch</b>	<b>0.000094'' (y<sup>+</sup> = 0.62)</b>	<b>0.000063'' (y<sup>+</sup> = 0.41)</b>	<b>0.000042'' (y<sup>+</sup> = 0.26)</b>

# Grid Information

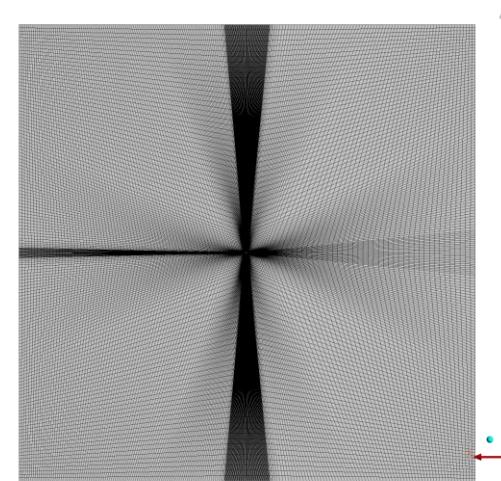
## Grid 1



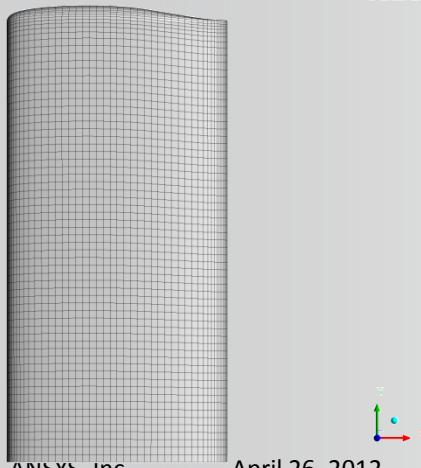
## Grid 2



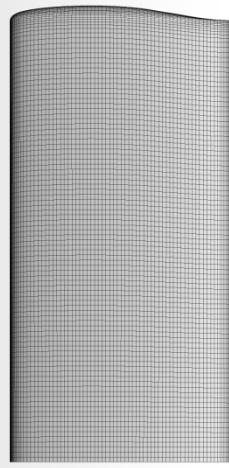
## Grid 3



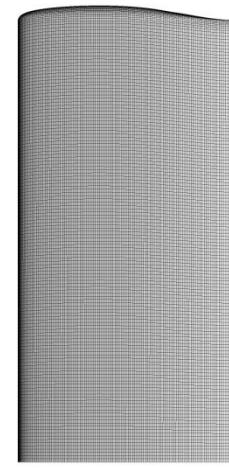
ANSYS



ANSYS

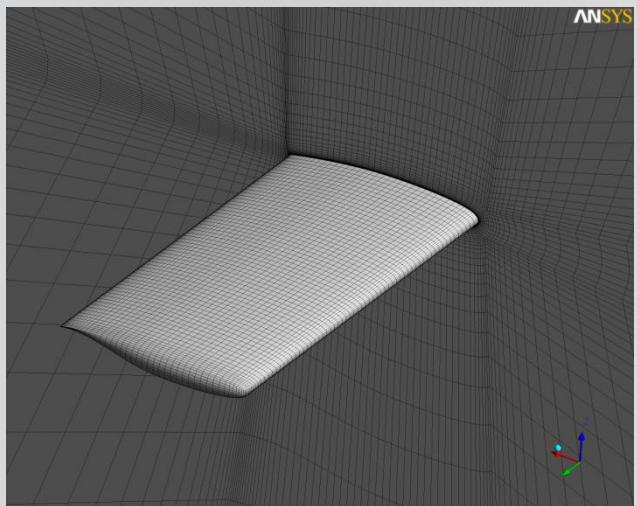


ANSYS

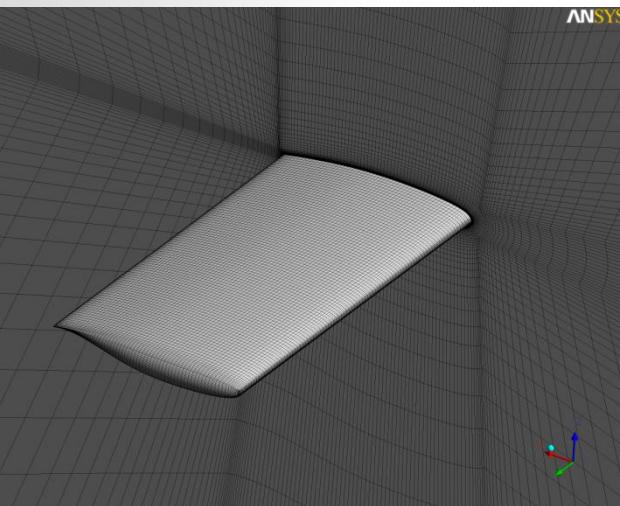


# Grid Information

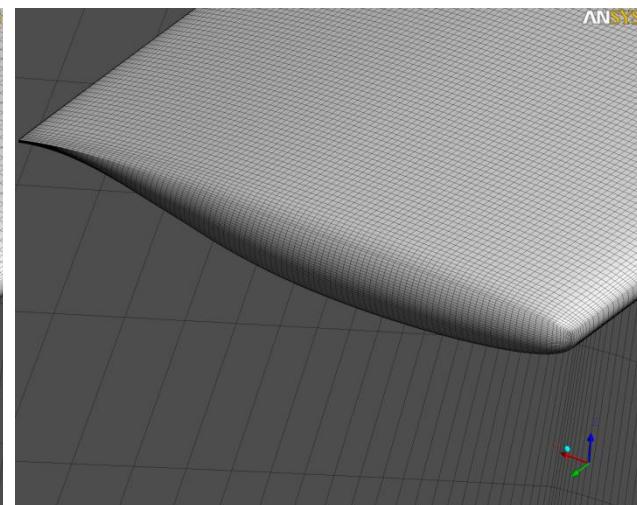
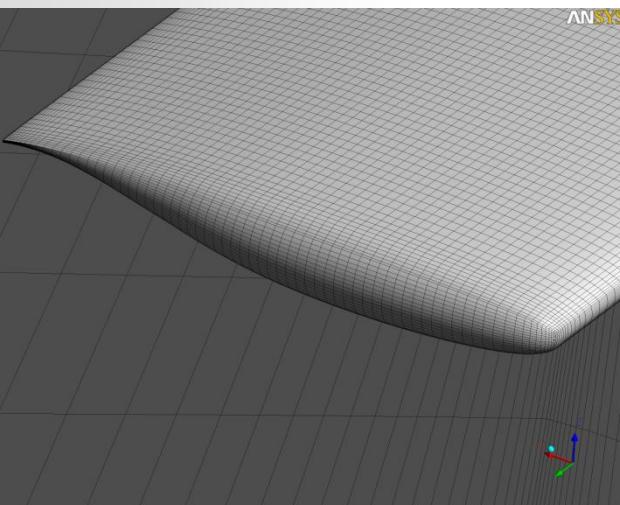
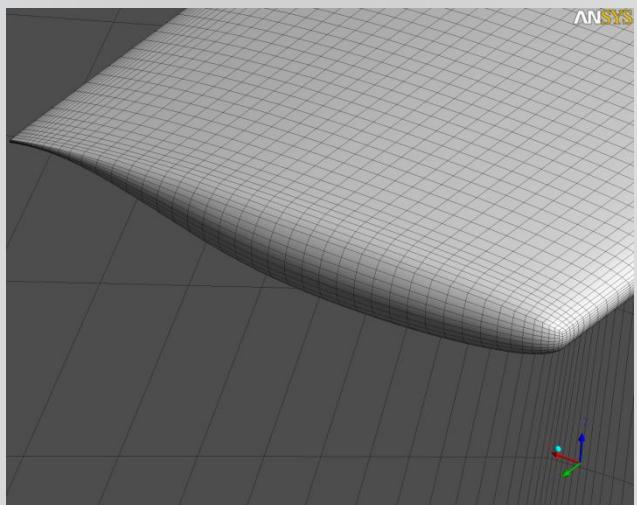
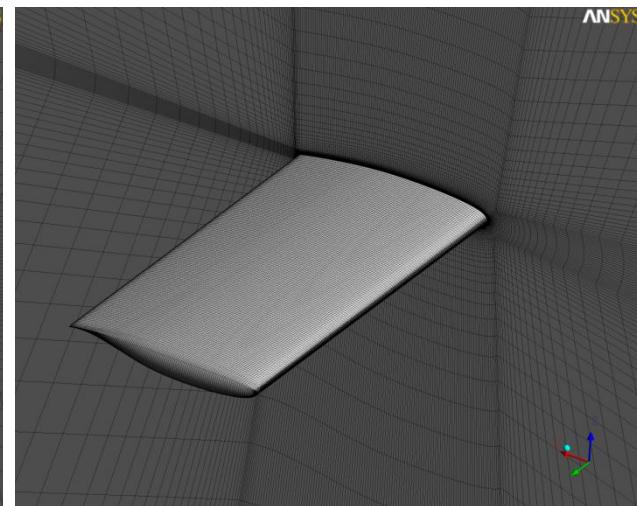
## Grid 1



## Grid 2

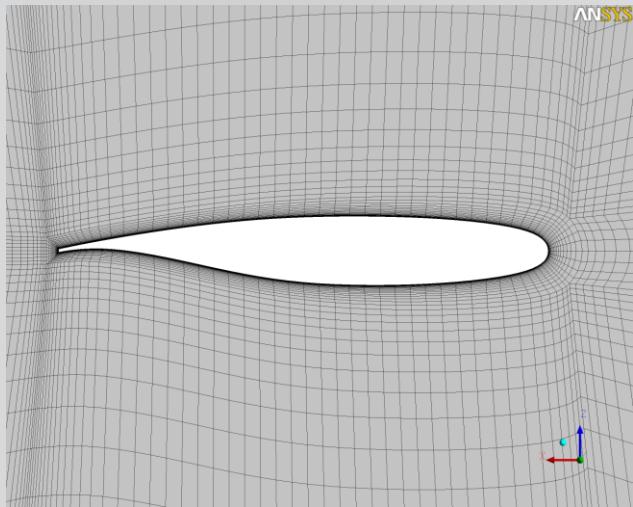


## Grid 3

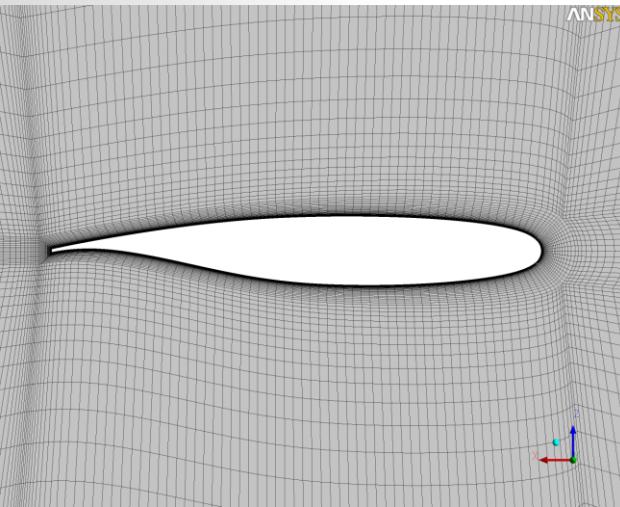


# Grid Information

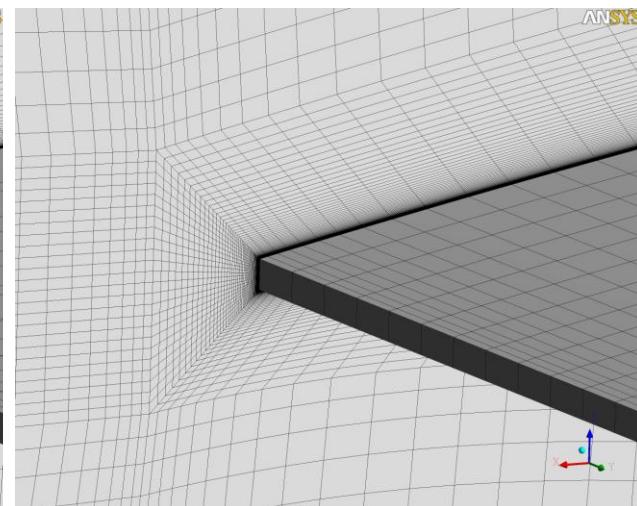
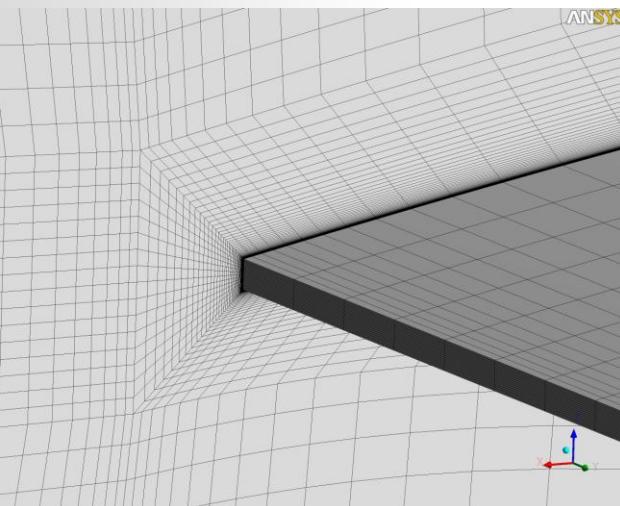
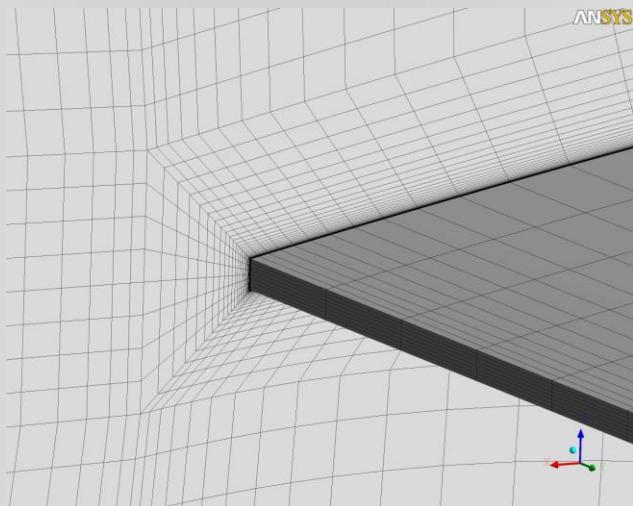
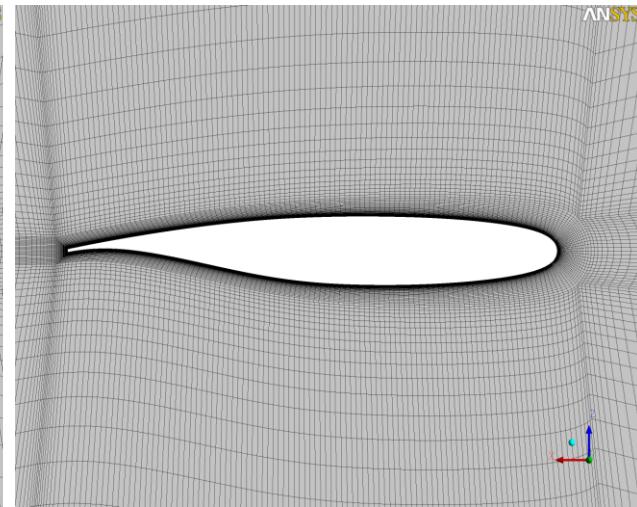
## Grid 1



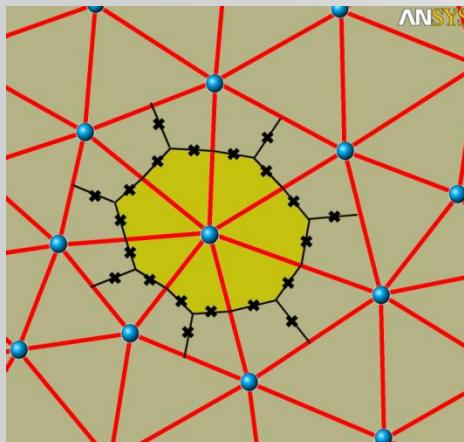
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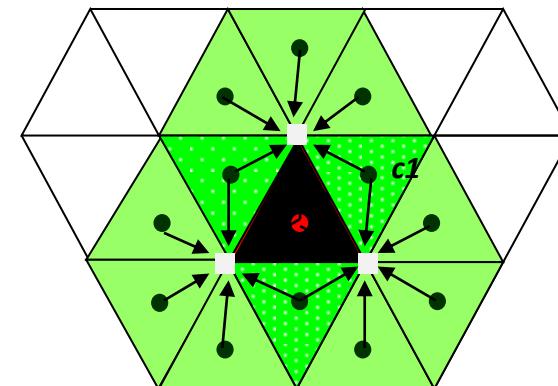
## Grid 3



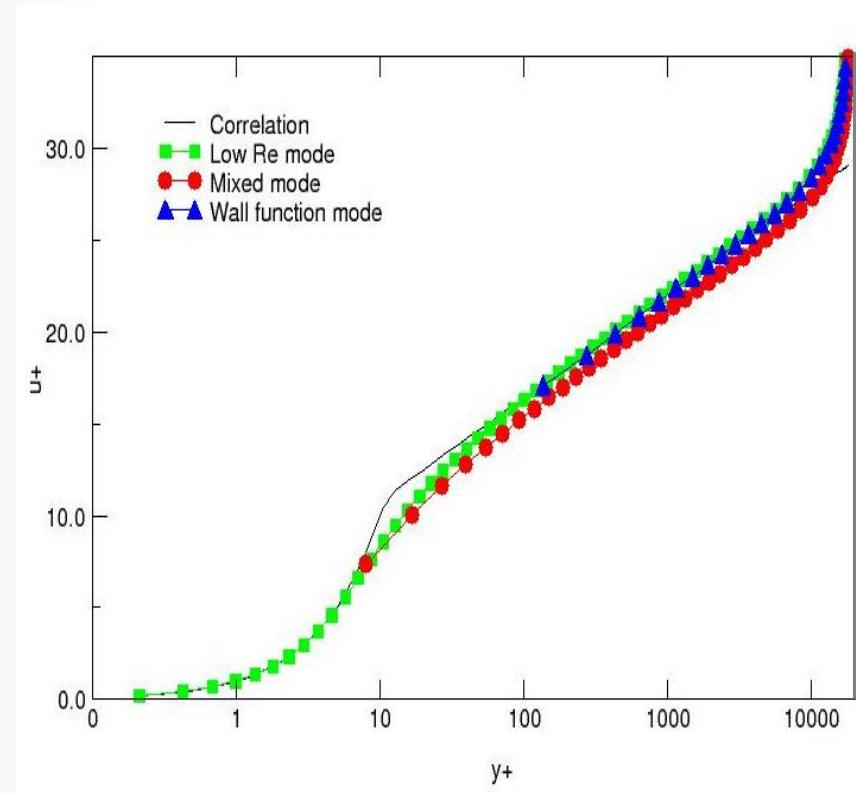
- ANSYS CFX 14
- Coupled (U,V,W,P) solver
  - Pressure based
- Convective discretization
  - High-resolution scheme
- Algebraic multigrid
- Vertex centred



- ANSYS FLUENT 14
- Coupled (U,V,W,P) solver
  - Pressure based
- Convective discretization
  - 2<sup>nd</sup> order upwind
- Algebraic multigrid
- Cell centred

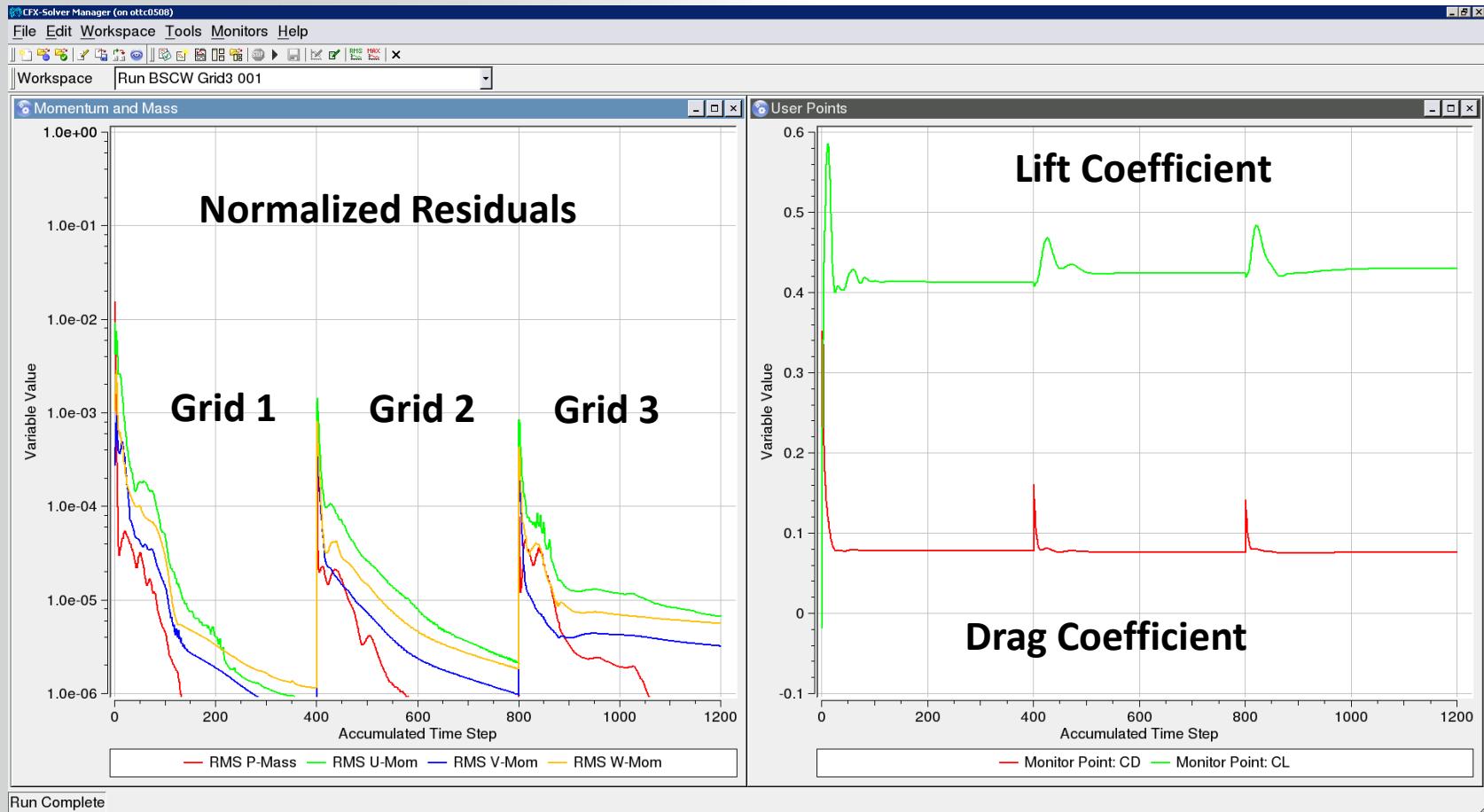


- Ensemble-averaged mass, momentum and energy conservation equations
- Turbulence model – SST (Menter, 1994)
- Automatic choice of linear/logarithmic near wall profiles



## Automatic Wall Treatment

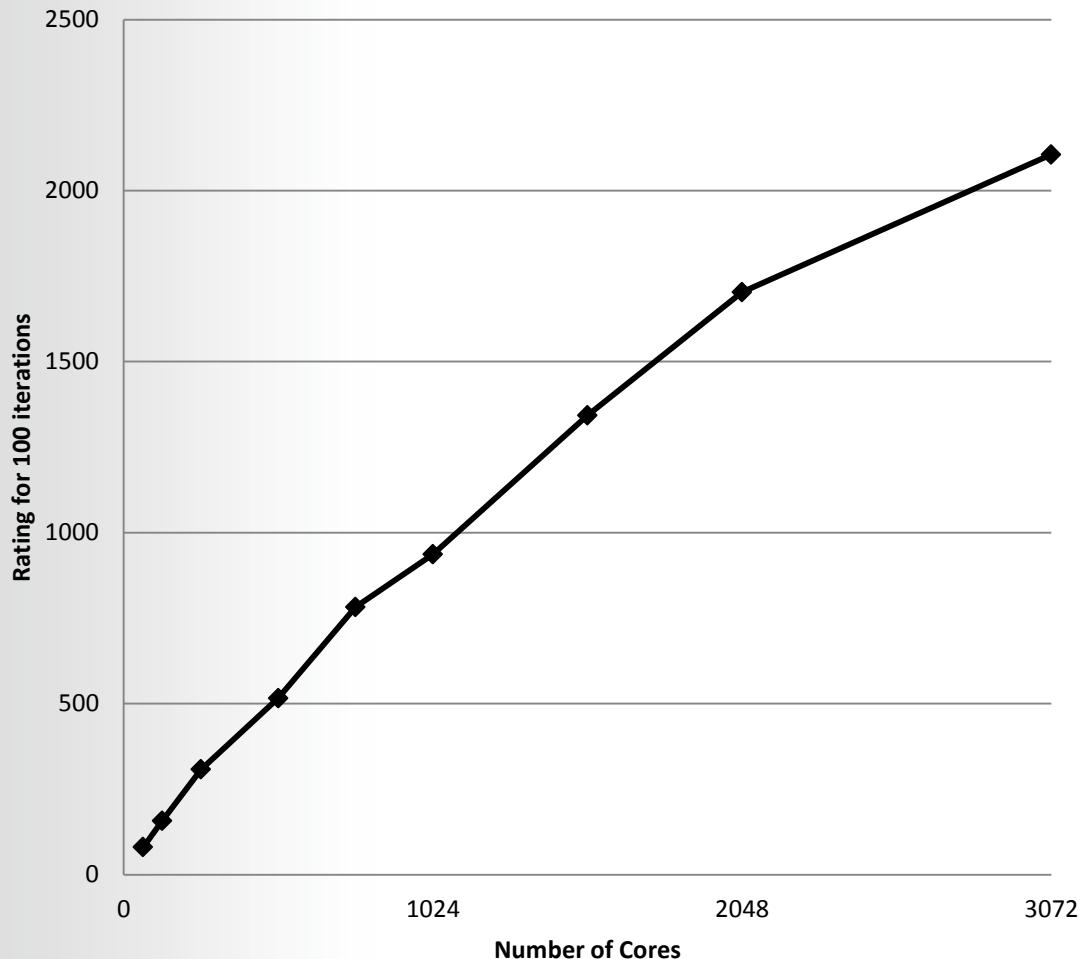
# CFX Solver, Steady-State



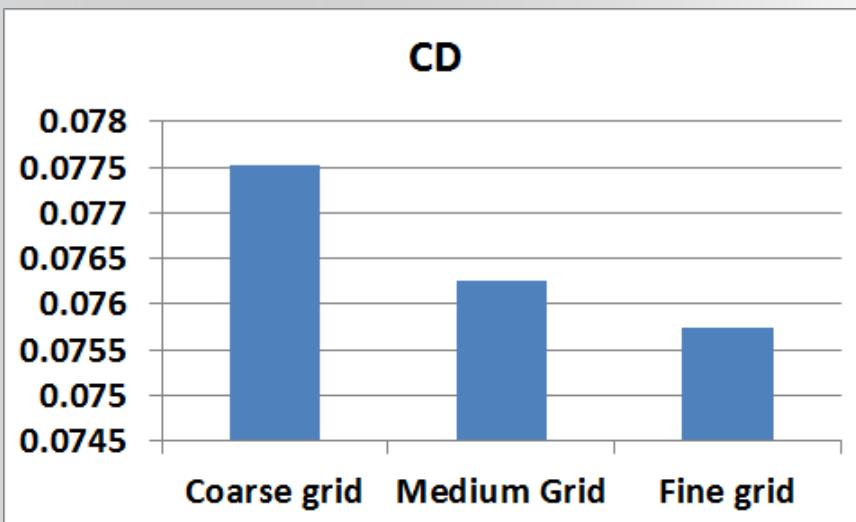
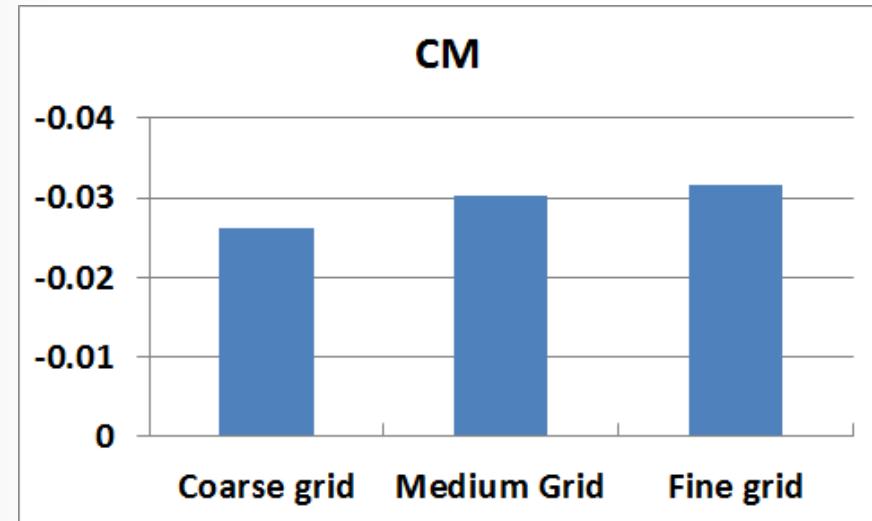
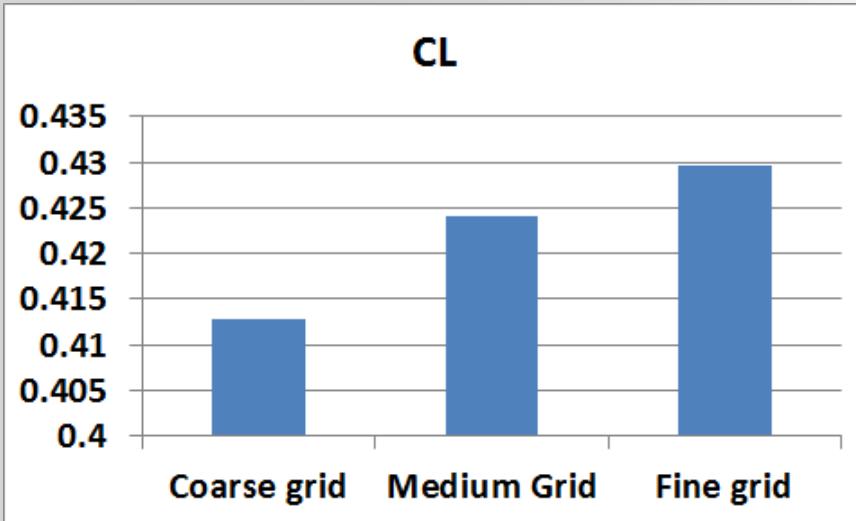
# CFX Solver, Steady-State

	# of elements, mio	# of iterations	# of CPUs	Total Wall Clock Time	Memory, Gbyte
Grid 1	1,45	400	36 14.5 % Ovlp	0 h 26 min	3.67
Grid 2	4,94	400	36 10.5 % Ovlp	1 h 21 min	7.61
Grid 3	13,75	400	36 7.8 % Ovlp	3 h 46 min	19.97

- **BSCW Grid-3**
  - 13.7 million cells
- **ANSYS Fluent**
- **CRAY XE6 Environment**
  - Cray MPI
  - Cray “Gemini” Interconnect
  - AMD “Interlagos”, 16 core/socket, 2.1GHz

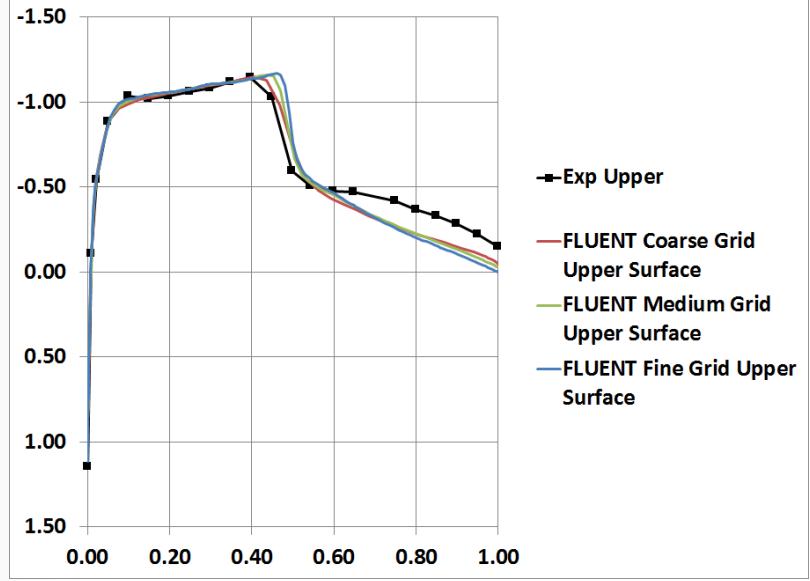
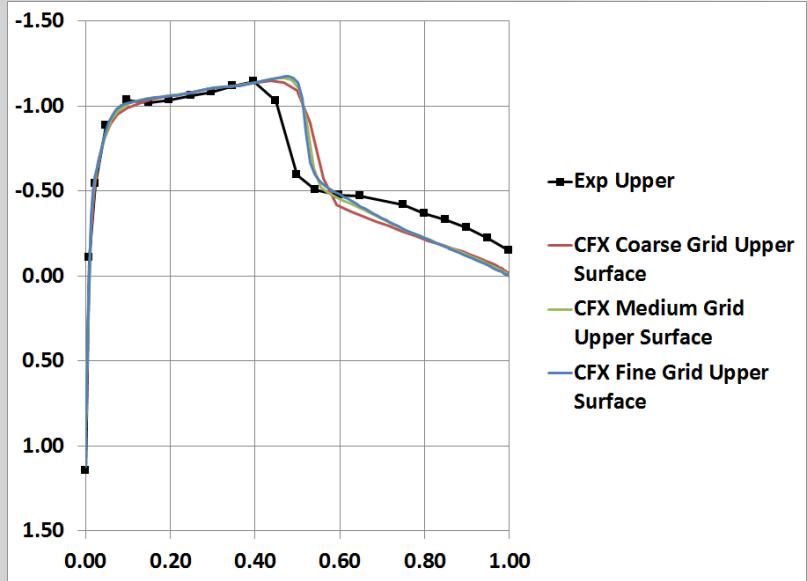
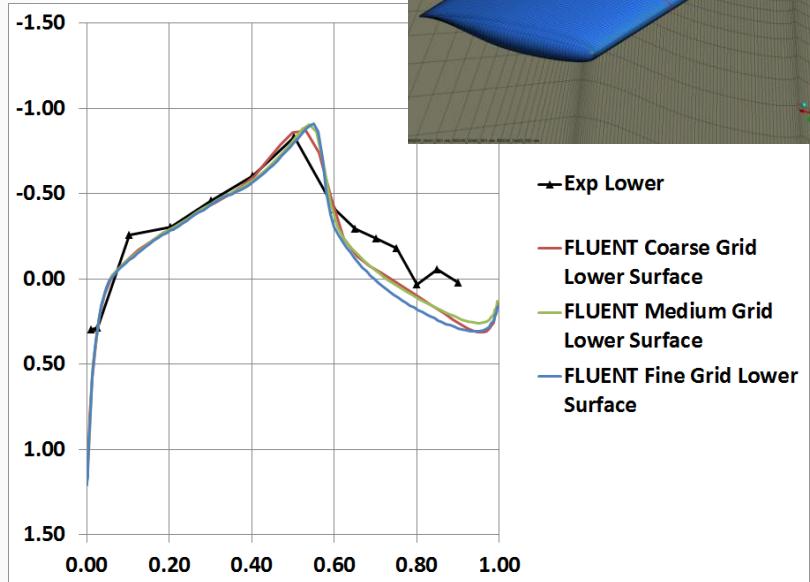
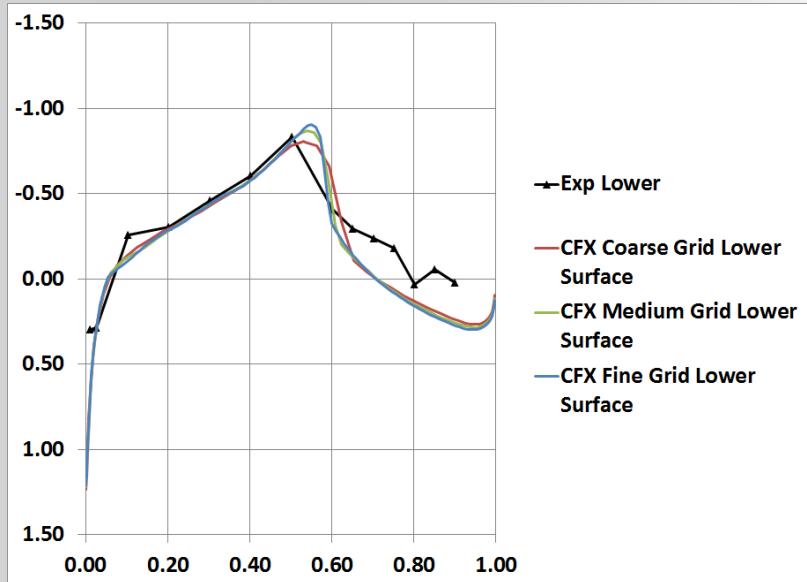
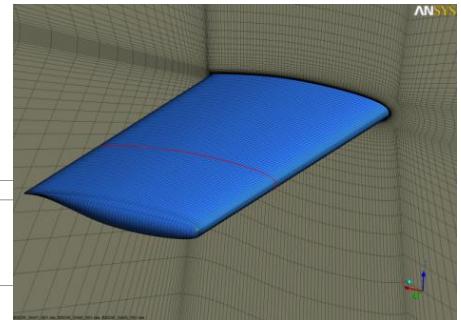


# ANSYS CFX: Lift, Moment, Drag



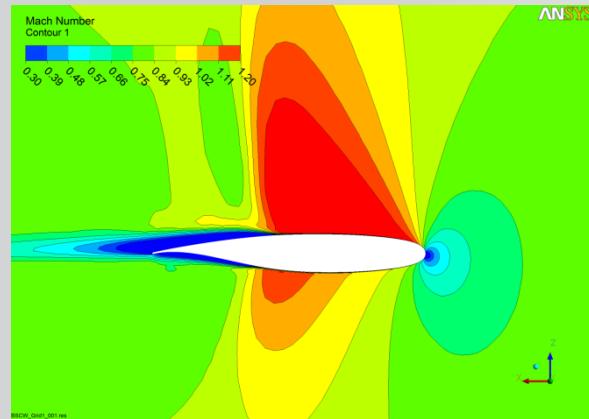
	CL	CM	CD
Grid 1	0.41277 (3.94 %)	-0.02630 (16.86 %)	0.07752 (2.34 %)
Grid 2	0.42417 (1.29 %)	-0.03020 (4.55 %)	0.07625 (0.67 %)
Grid 3	0.42972	-0.03164	0.07574

# BSCW, CP @ Eta = 60 %

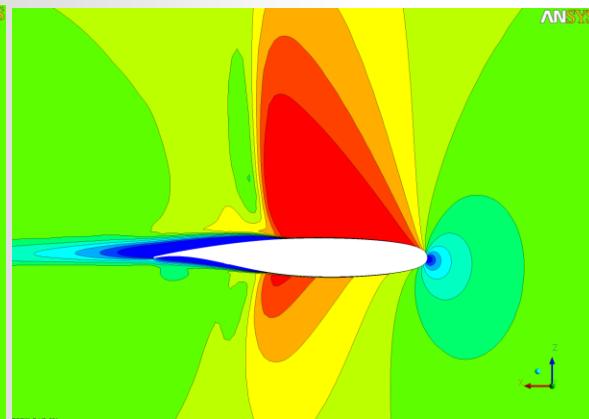


# BSCW, Mach Number @ Eta = 60 %

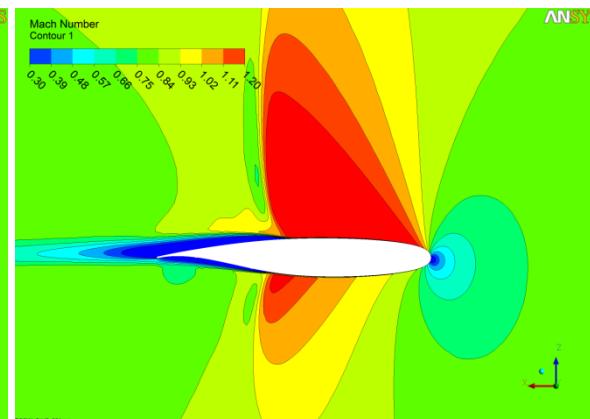
Grid1



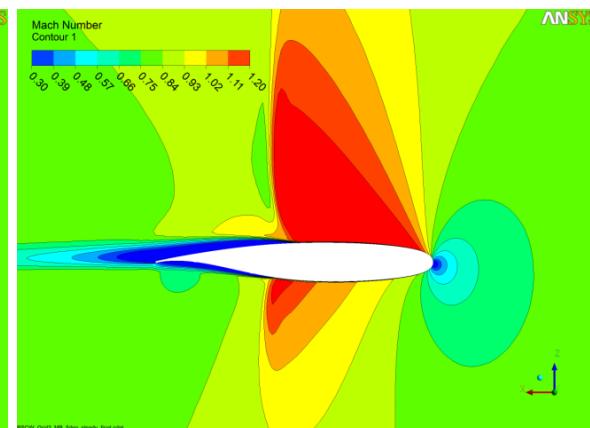
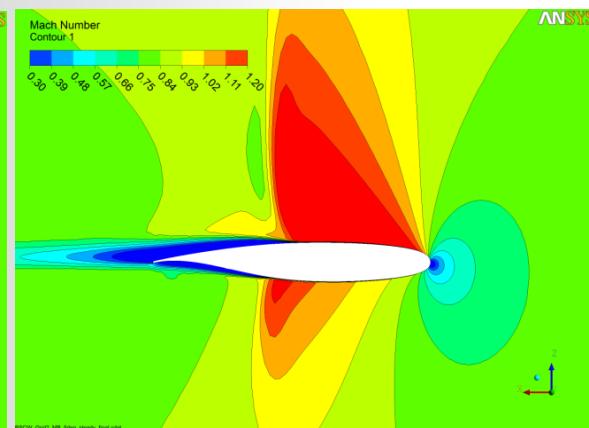
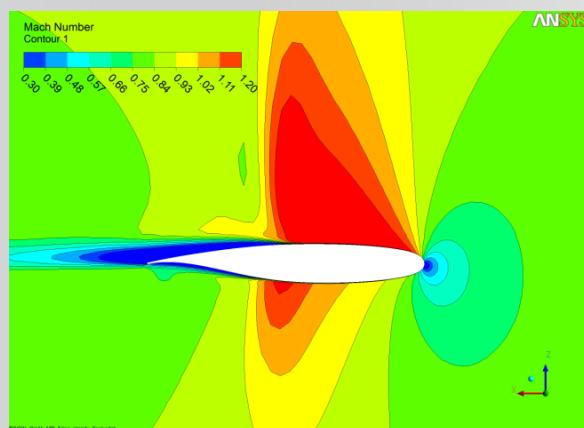
Grid2



Grid 3

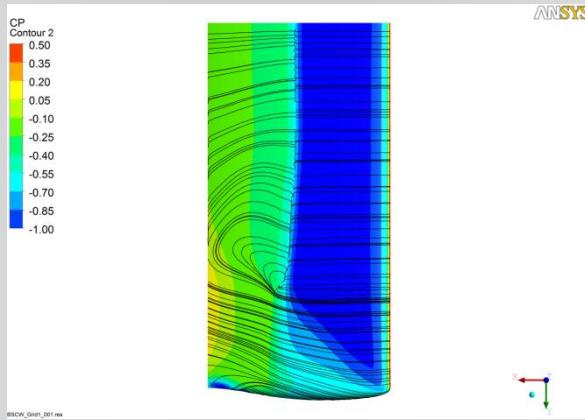


FLUENT

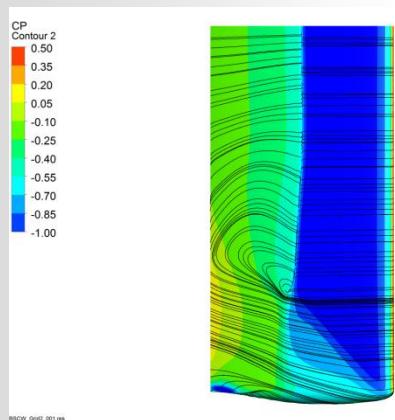


# BSCW, CP @ Wing Surface, CFX

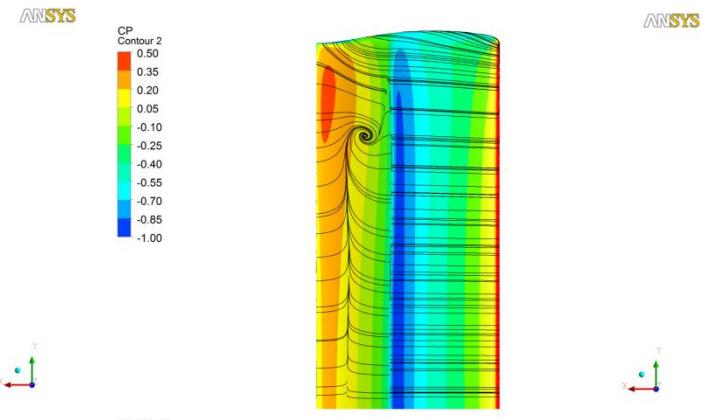
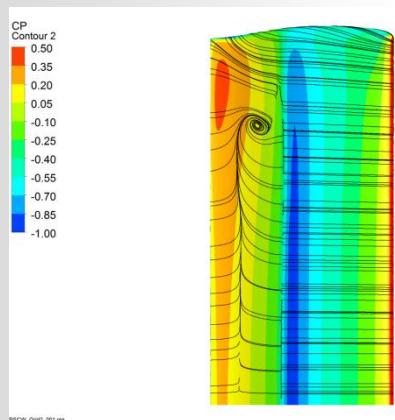
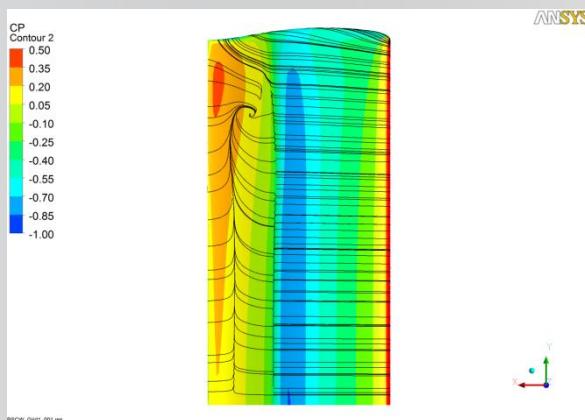
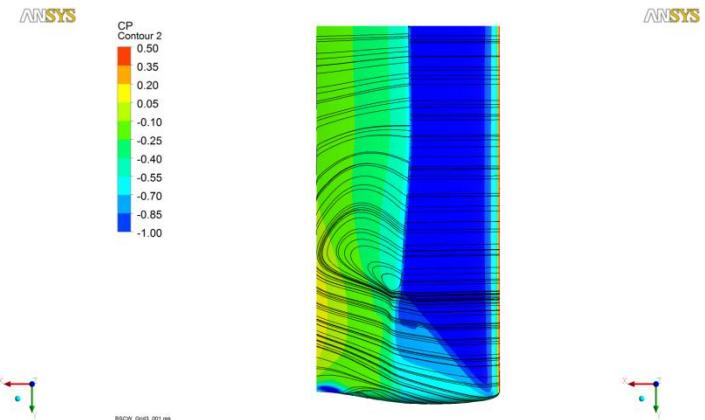
## Grid1



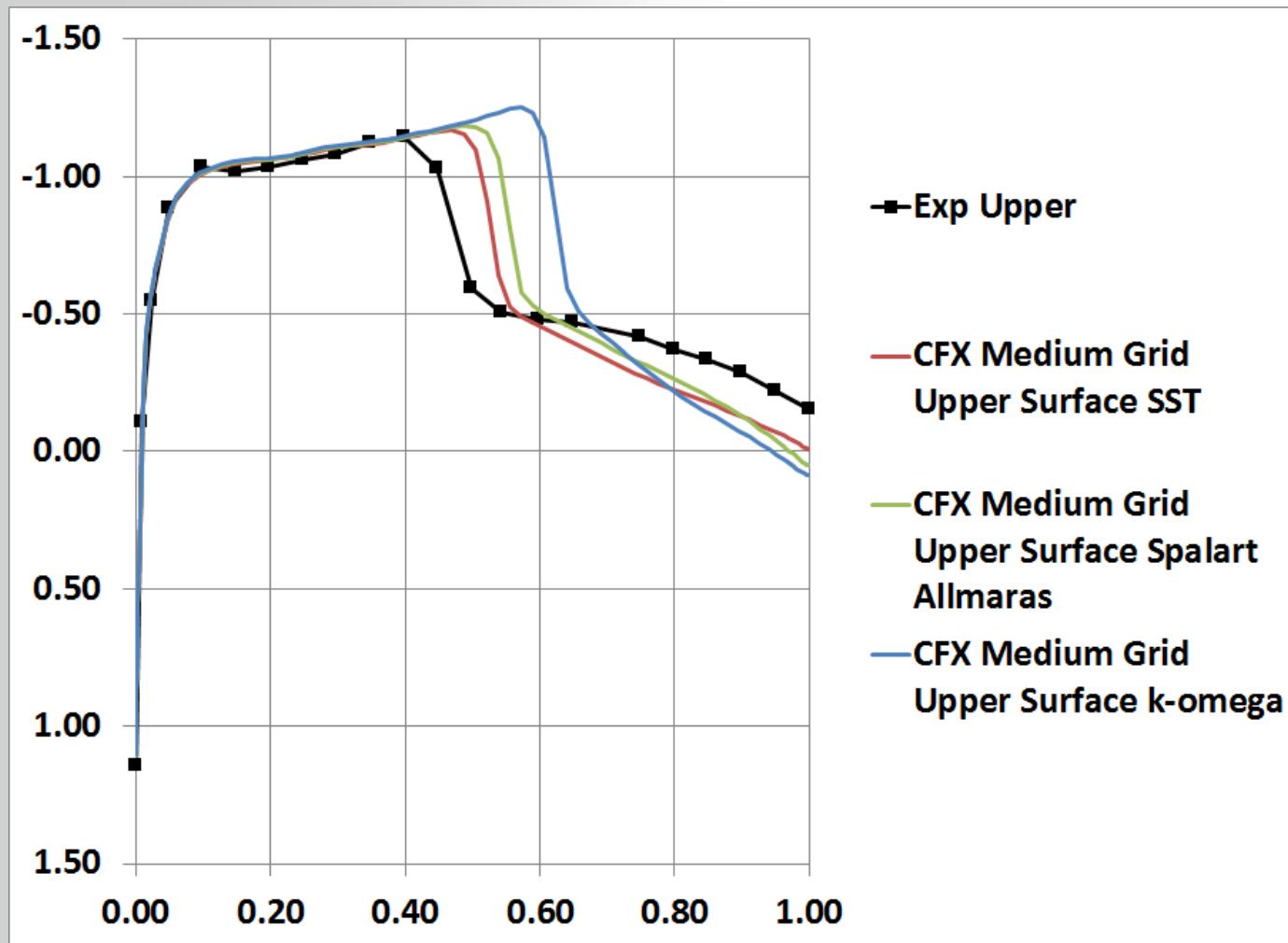
## Grid2



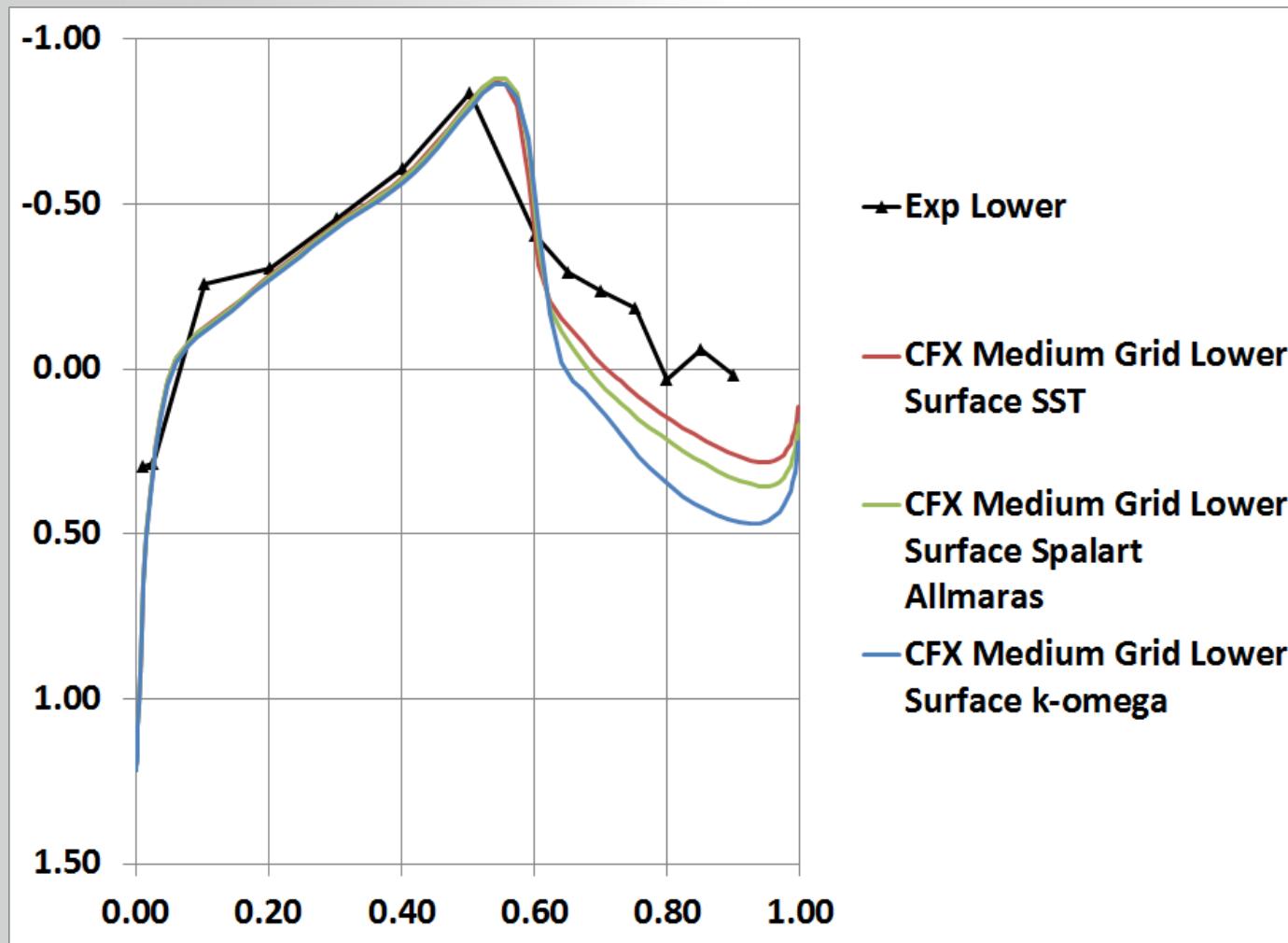
## Grid 3



# Turbulence Model Error Estimation

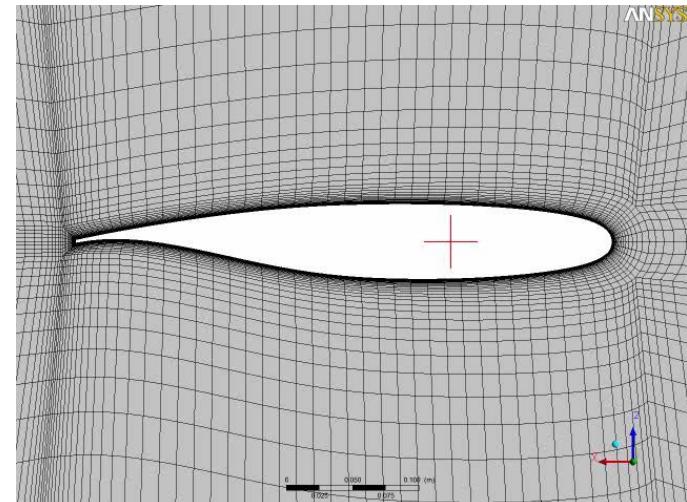
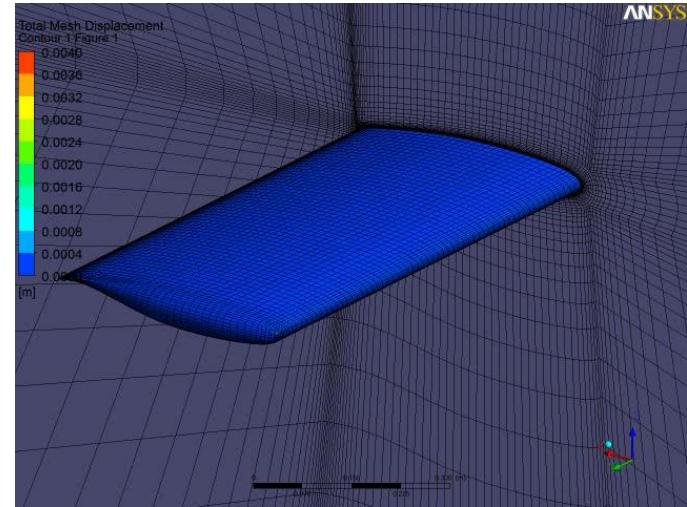


# Turbulence Model Error Estimation



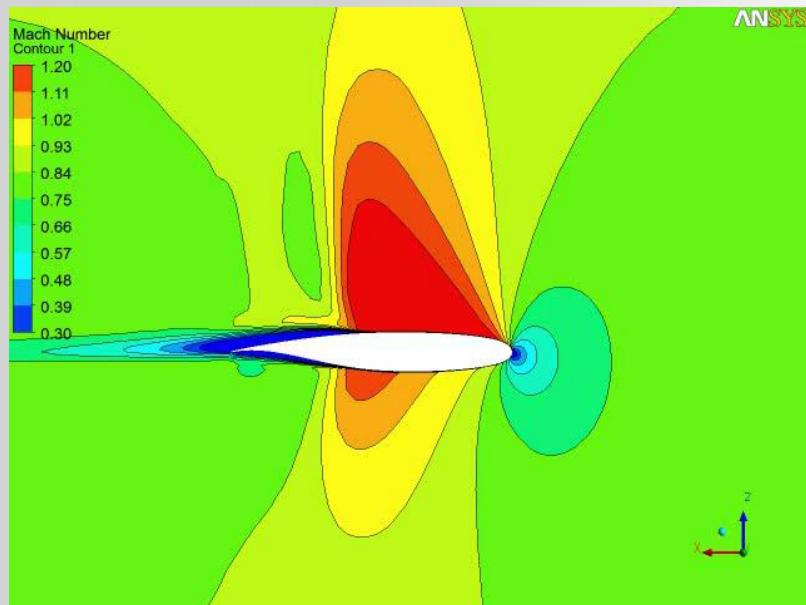
# Unsteady-State Calculation

- Pitching oscillations
- Mesh displacement
  - Harmonic wing motion
  - $A \cdot \sin(\omega \cdot t)$
- Initial condition
  - Converged steady-state solution
- Monitor frequencies
- FFT

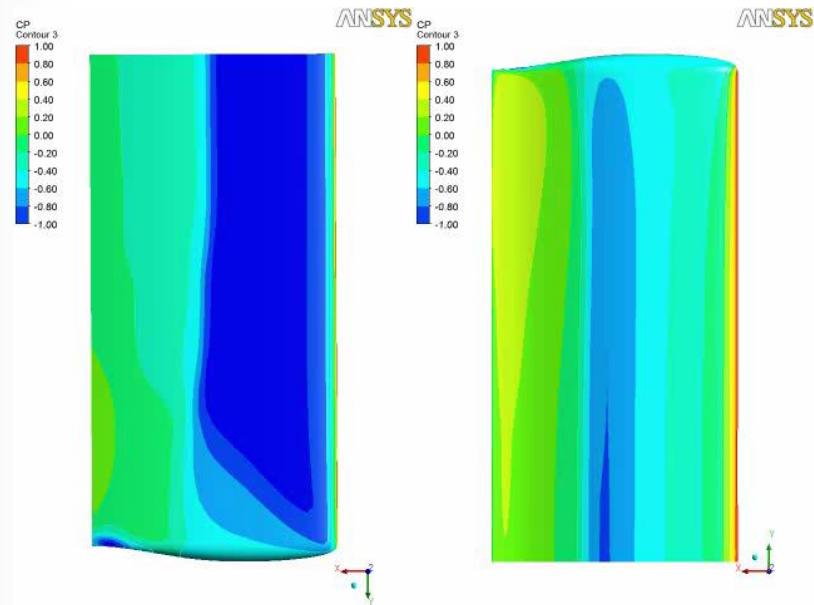


# Animation

Mach Number @  $\eta = 60\%$



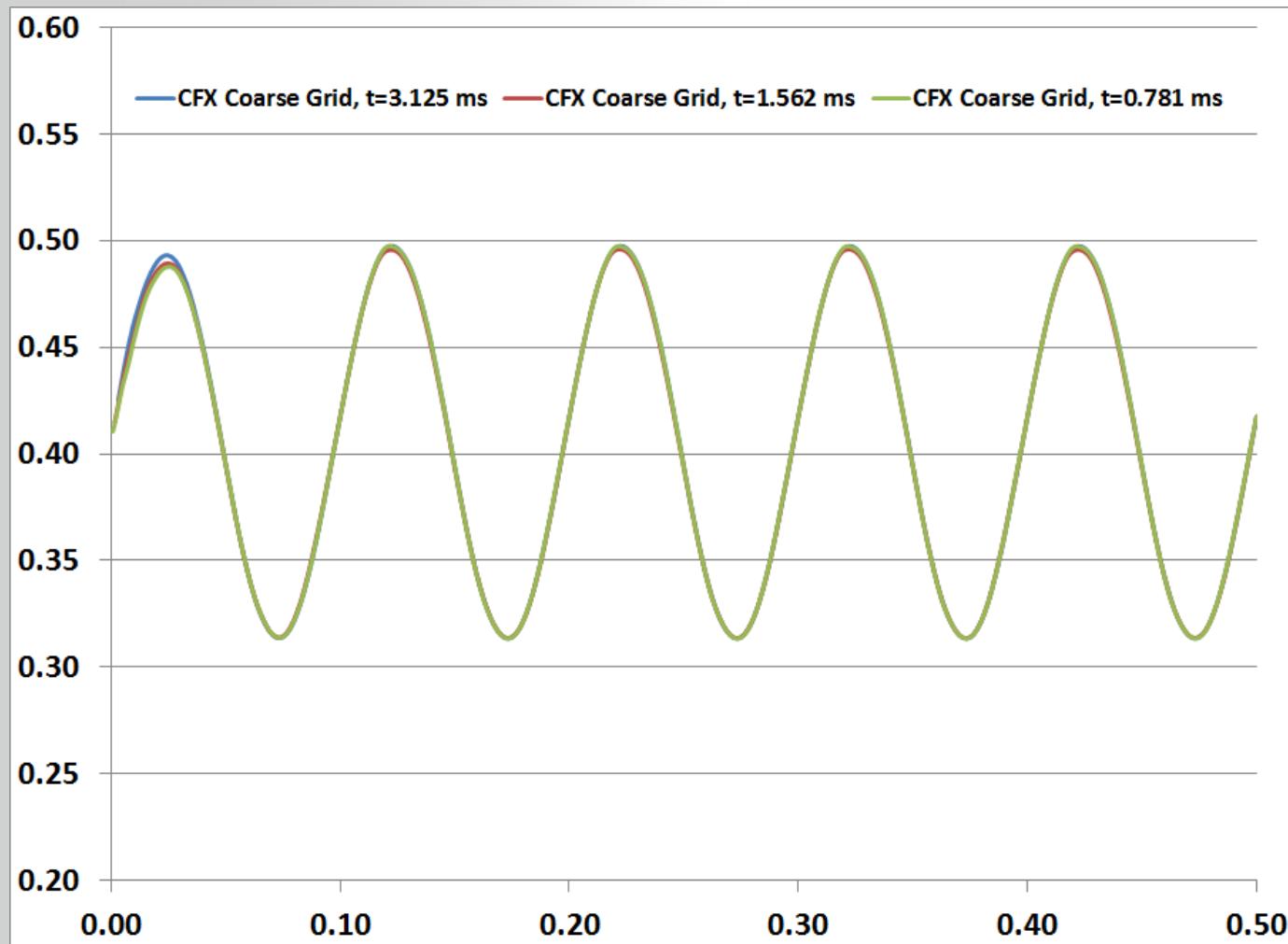
CP @ Lower & Upper Surface



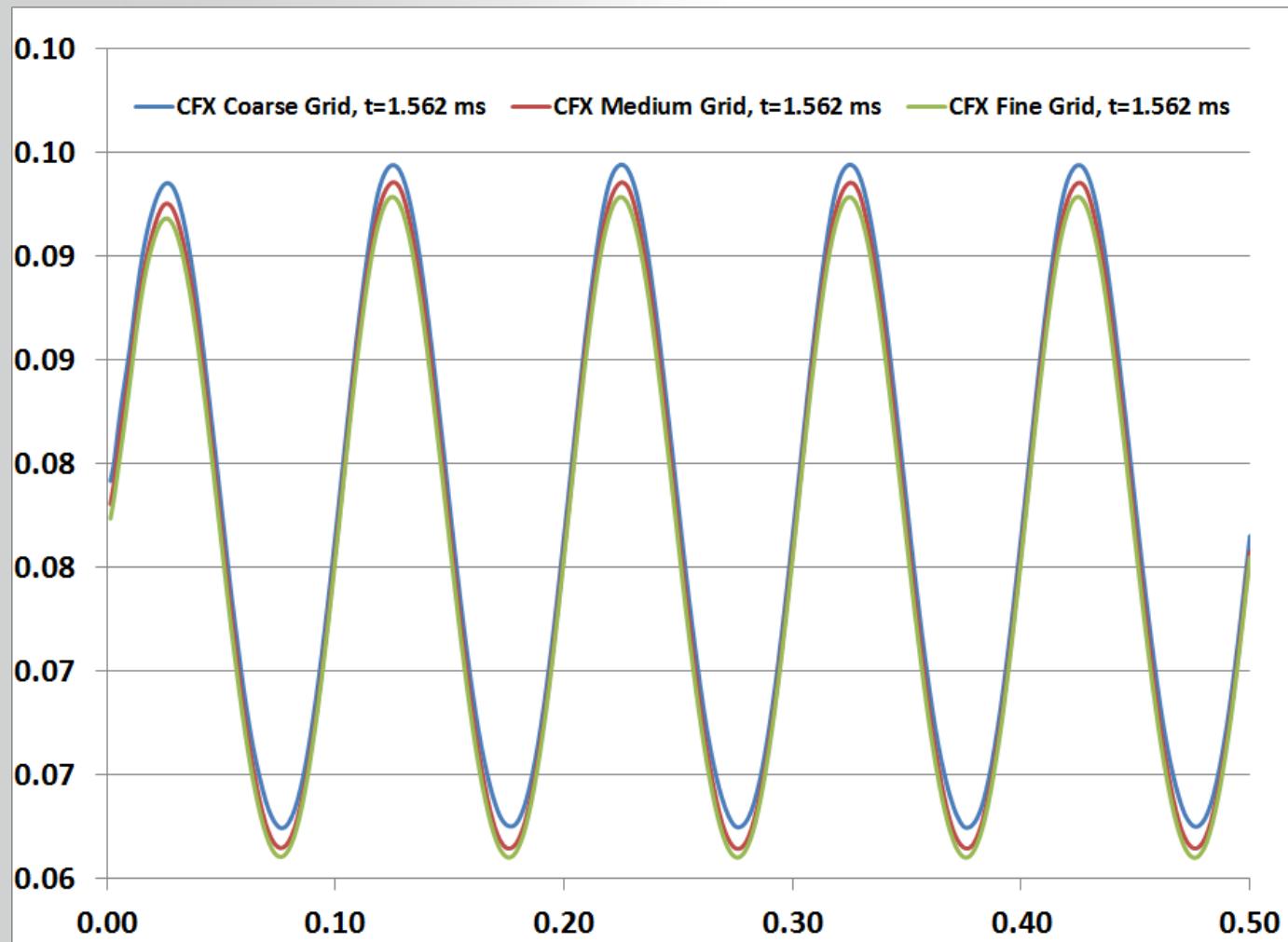
# Numerical Information

- Transient scheme
  - Second order backward Euler
- Convective discretization
  - High Resolution
- Initial condition
  - Steady-state solution
- Time steps per period
  - Run1: **32** > 3.125 ms
  - Run2: **64** > 1.562 ms
  - Run3: **128** > 0.781 ms
- Total time = 5 \* period
  - $32 * 5 = 160$  iterations
  - $64 * 5 = 320$  iterations
  - $128 * 5 = 640$  iterations

# Lift Coefficient, 10 Hz, Temporal Error



# Drag Coefficient, Spatial Error



# Frequency Response Function

- Fourier Series is written in form of sines and cosines

$$x(t) = a_0 + \sum_{n=1}^{\infty} a_n \cos(n\omega_0 t) + b_n \sin(nw_0 t)$$

- where for  $n > 0$ :

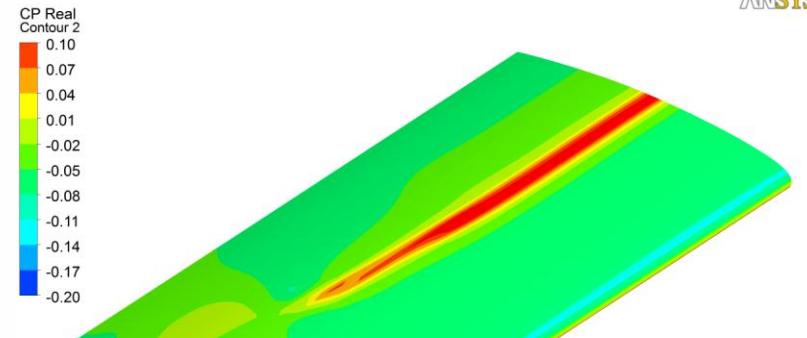
$$a_n = \frac{2}{T} \int_{t_0}^{t_0+T} x(t) \cos(nw_0 t) dt, \quad b_n = \frac{2}{T} \int_{t_0}^{t_0+T} x(t) \sin(nw_0 t) dt$$

- and where

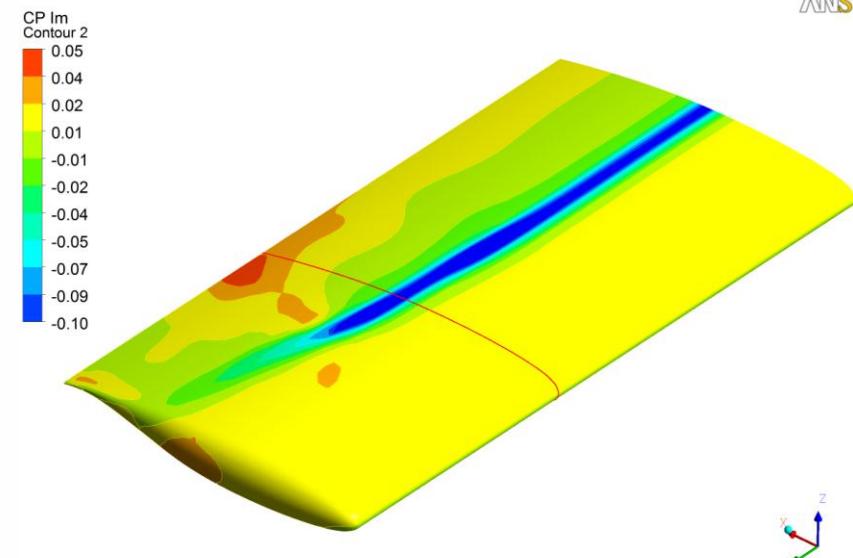
$$\omega_0 = \frac{2\pi}{T}, \quad a_0 = \frac{1}{T} \int_{t_0}^{t_0+T} x(t) dt$$

# Frequency Response Function

- Fourier coefficients calculated in CFD solver

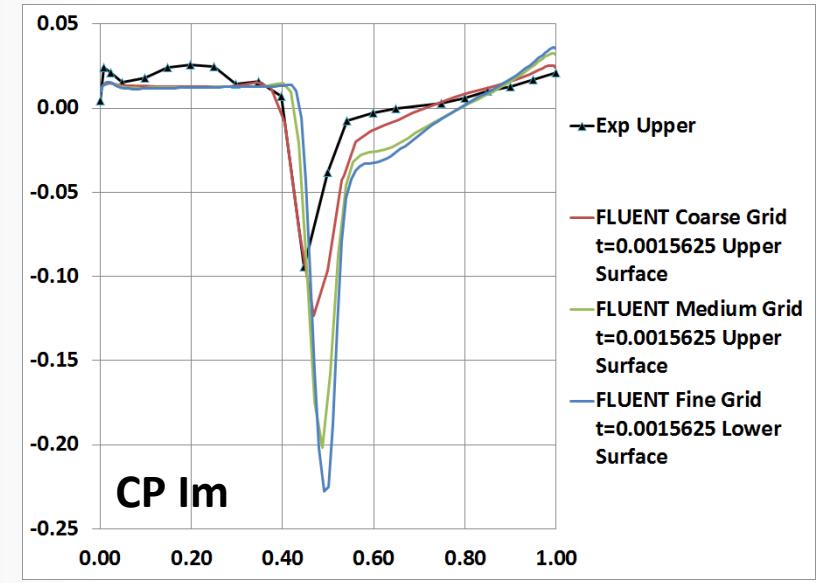
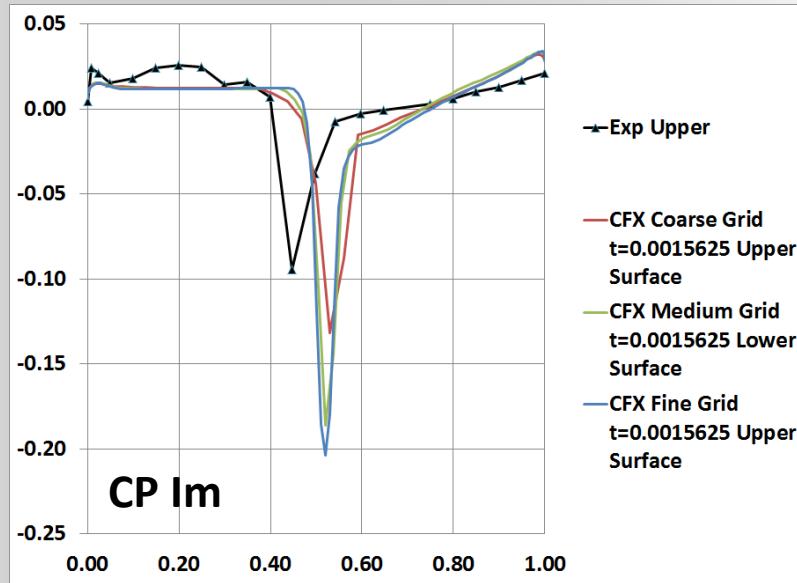
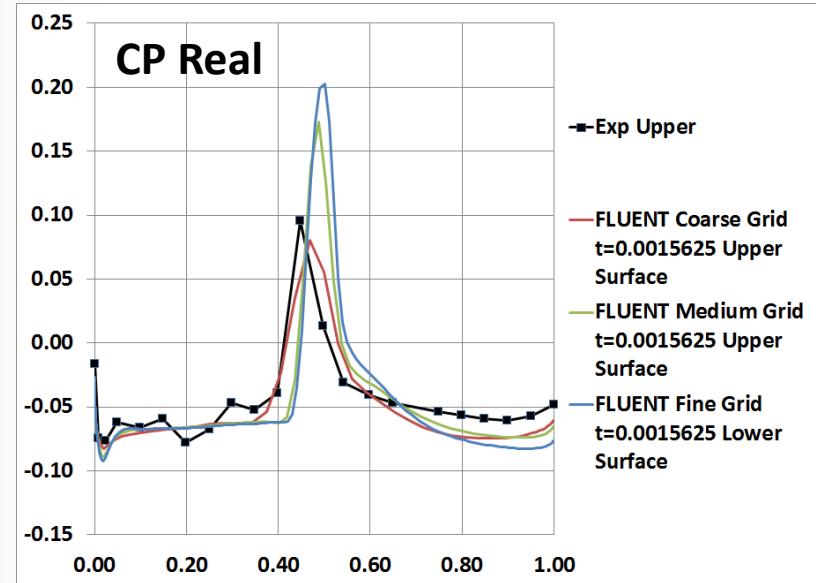
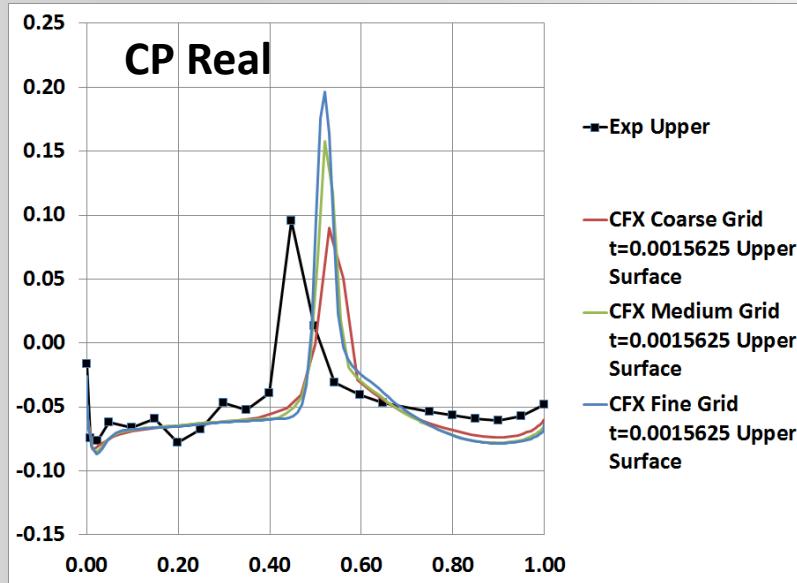


- CFX
  - Additional Variable
    - CP Real
    - CP Im
    - Magnitude =  $\sqrt{CP\ Real^2+CP\ Im^2}$
    - Phase =  $\text{atan2}(CP\ im/CP\ Real)$

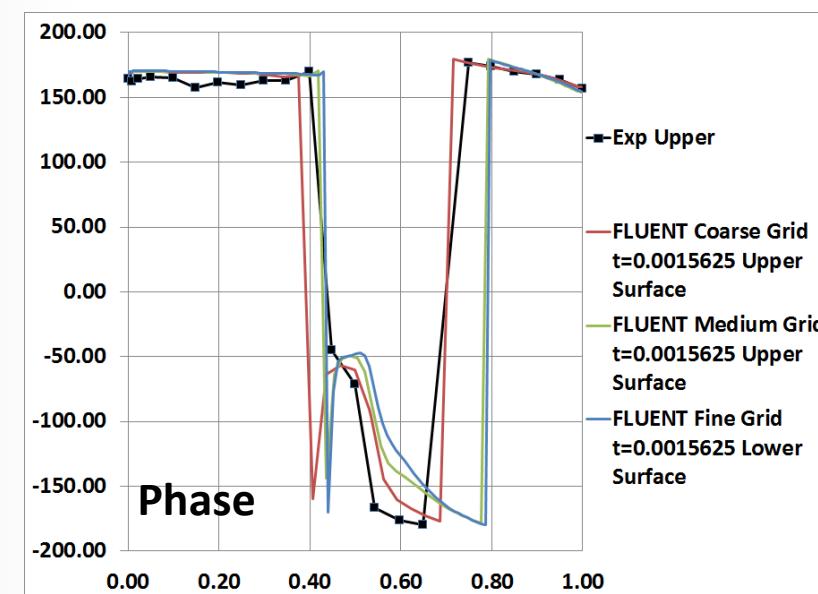
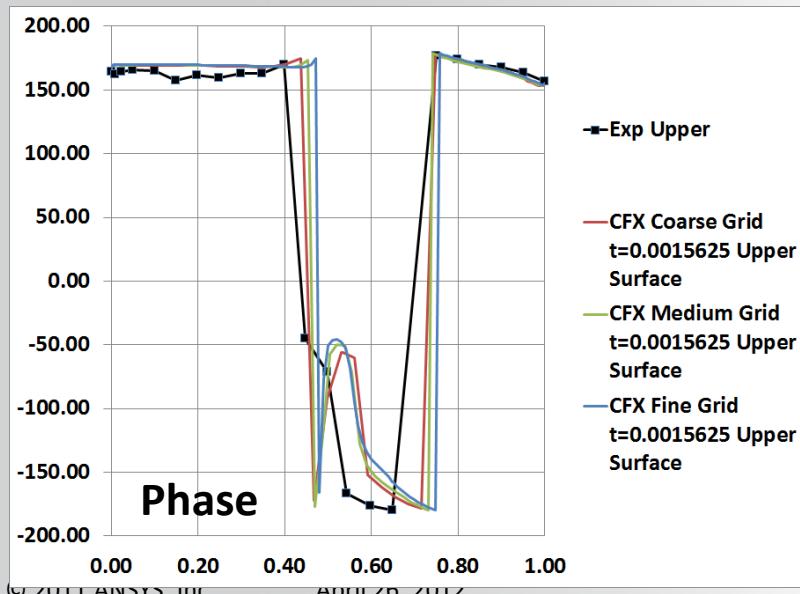
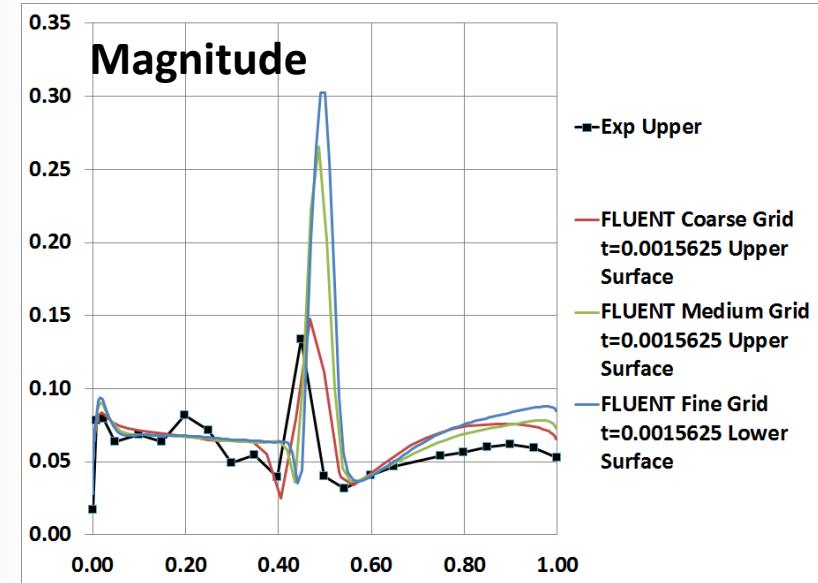
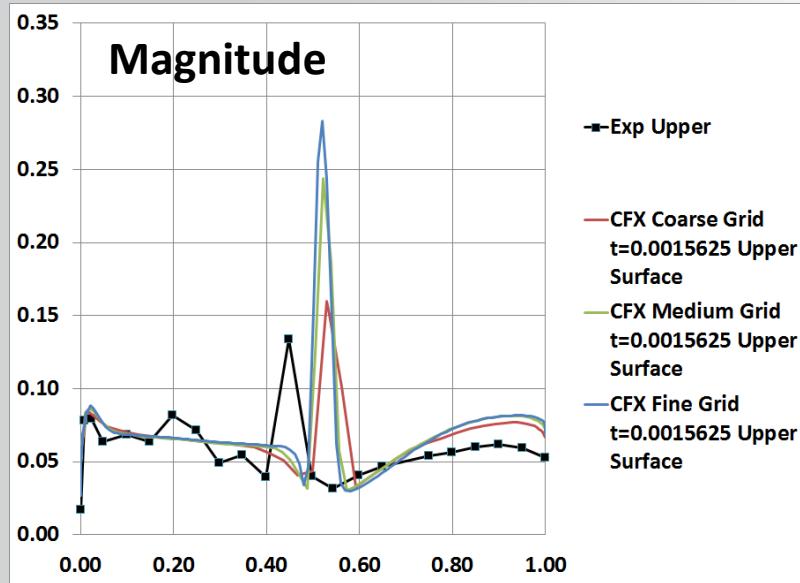


- FLUENT
  - User Defined Function
    - CP Real
    - CP Im
    - Magnitude =  $\sqrt{CP\ Real^2+CP\ Im^2}$
    - Phase =  $\text{atan2}(CP\ im/CP\ Real)$

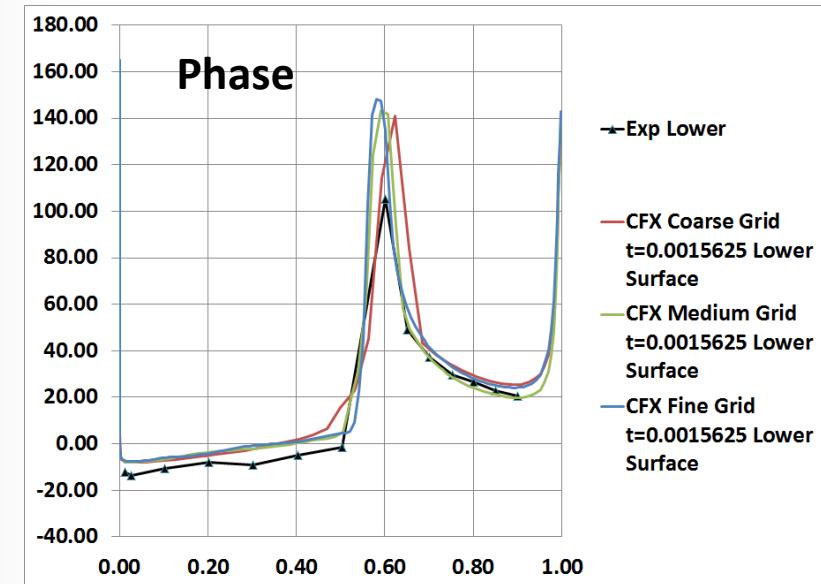
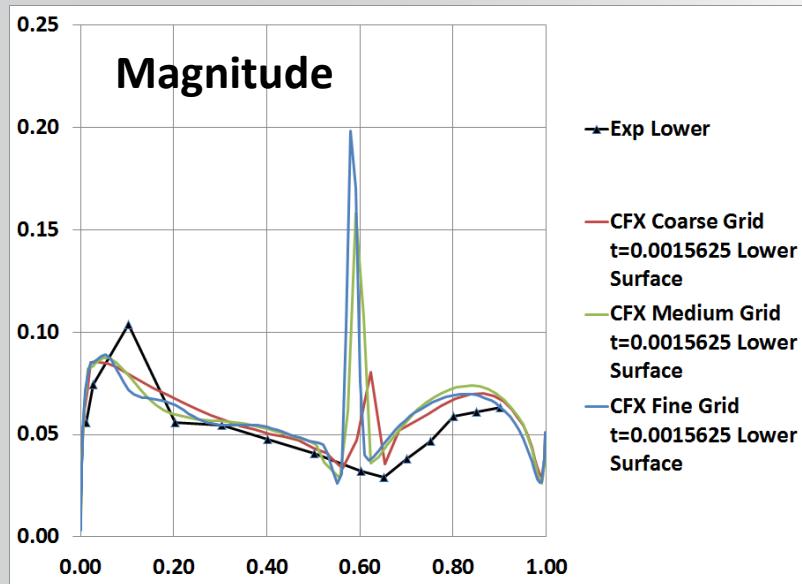
# BSCW 10 Hz @ Eta=0.6, Upper



# BSCW 10 Hz @ Eta=0.6, Upper



# BSCW 10 Hz @ Eta=0.6, Lower



# Summary & Outlook

- ANSYS CFD calculation of a forced oscillation pitching motion of a Supercritical I-Wing
- Detailed quality assurance of numerical errors
  - Iteration error
  - Discretization error (Spatial and temporal)
  - Model error